

Amendments to the Claims:

1. (Currently Amended) A missile comprising:  
a fuselage member configured to carry an engine;  
a wing actuator carried by the fuselage member; and  
an oblique wing member pivotally connected to the fuselage member on a lower surface thereof, wherein the wing member comprises a chord length that is less than a diameter of the fuselage member, and wherein the wing member is pivotable by the wing actuator from a stationary position substantially aligned with the fuselage member at less than transonic speed during flight while the engine is not initiated to a predetermined sweep angle of less than 90 degrees at transonic speed during flight.
2. (Original) A missile according to Claim 1, wherein the wing member has an aspect ratio of less than about 7.0.
3. (Canceled).
4. (Canceled).
5. (Original) A missile according to Claim 1, wherein the wing member is pivoted to a sweep angle of approximately 30 to 40 degrees such that one end of the wing extends generally in the direction of flight and an opposite end generally trails behind.
6. (Original) A missile according to Claim 1, wherein the wing member is pivotally attached to the fuselage member proximate to the midpoint of the wing member.
7. (Original) A missile according to Claim 1, wherein the wing member is pivotally attached to the fuselage member proximate to a one-quarter chord of the wing member.
8. (Original) A missile according to Claim 1, wherein the fuselage member has a diameter of less than about 7 inches.

9. (Original) A missile according to Claim 1, wherein the missile may thrust to transonic speeds of at least about Mach 0.9.

10. (Original) A missile according to Claim 1, wherein the missile may thrust to and maintain transonic speeds for at least 30 minutes during flight.

11. (Original) A missile according to Claim 1, further comprising a restraint attached to the fuselage and positioned proximate to each end of the wing member such that when the wing member is substantially aligned with the fuselage member, the wing member is substantially free from vibration.

12. (Original) A missile according to Claim 1, further comprising an antenna positioned within, and substantially along, an entire length of the wing member.

13. (Original) A missile according to Claim 1, wherein the wing actuator comprises a wound, spring-loaded actuator.

14. (Withdrawn) A missile according to Claim 1, wherein the wing actuator comprises an electronic actuator that may vary the sweep angles during flight.

15. (Original) A missile according to Claim 1, further comprising fins pivotally attached and proximate to a trailing end of the fuselage member.

16. (Currently Amended) A missile system comprising a missile configured to be releasably attached to an aircraft, the missile comprising:  
a fuselage member configured to carry an engine;  
a wing actuator carried by the fuselage member; and  
an oblique wing member pivotally connected to the fuselage member on a lower surface of the fuselage member, wherein the wing member comprises a chord length that is less than a

diameter of the fuselage member, and wherein the wing member is pivotable by the wing actuator from a stationary position substantially aligned with the fuselage member at less than transonic speed during flight while the engine is not initiated to a predetermined sweep angle of approximately 30 to 40 degrees at transonic speed during flight.

17. (Original) A missile according to Claim 16, wherein the wing member has an aspect ratio of less than about 7.0.

18. (Canceled).

19. (Canceled).

20. Canceled.

21. (Original) A missile system according to Claim 16, wherein the missile may thrust to transonic speeds of at least about Mach 0.9.

22. (Original) A missile system according to Claim 16, wherein the missile may thrust to and maintain transonic speeds for at least 30 minutes.

Claims 23-28 (Canceled)

29. (New) The missile of Claim 1, wherein a ratio of the diameter of the fuselage to the chord length of the wing member is about 3:2.

30. (New) The missile of Claim 1, wherein a ratio of a length of the fuselage to a span of the wing member is about 7:2.